

XFOIL
V 6.96

C_p
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0

S1223

$Re = 0.200 \times 10^6$

$\alpha = 12.5000^\circ$

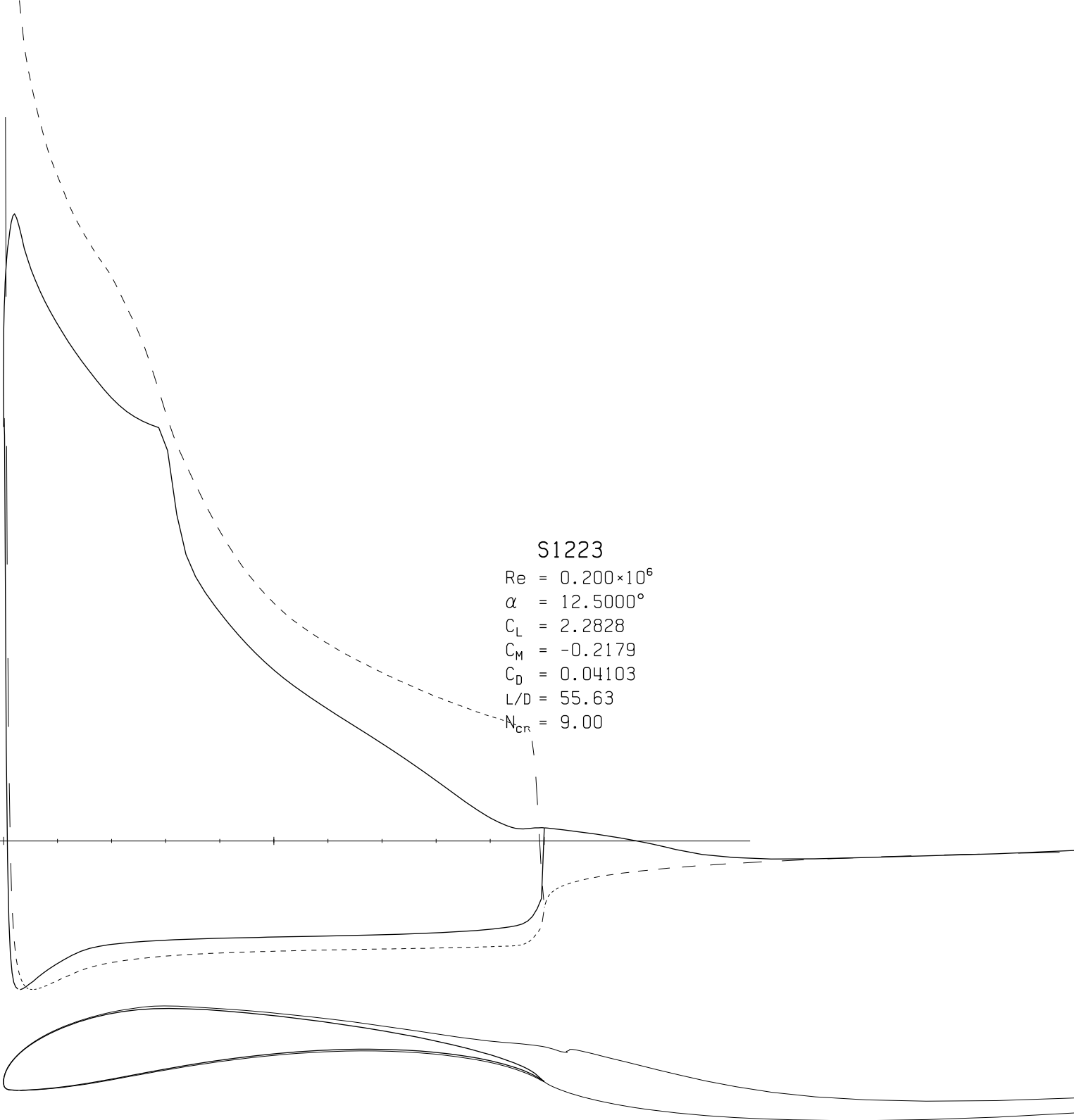
$C_L = 2.2828$

$C_M = -0.2179$

$C_D = 0.04103$

$L/D = 55.63$

$N_{cr} = 9.00$



S1223
area = 0.06491
thick. = 0.12141
camber = 0.08692
 r_{LE} = 0.03158
 $\Delta\theta_{TE}$ = 3.23°

